



US Department
of Transportation
Federal Aviation
Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved
OMB No. 2120-0020
11/30/2007

Electronic Tracking Number

For FAA Use Only

INSTRUCTIONS: Print or type all entries. See Title 14 CFR §43.9, Part 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. §44701). Failure to report can result in a civil penalty for each such violation. (49 U.S.C. §46301(a))

1. Aircraft	Nationality and Registration Mark	Serial No.		
	Make	Model	Series	
2. Owner	Name (As shown on registration certificate)		Address (As shown on registration certificate)	
	Address		City _____ State _____	
	Zip _____		Country _____	

3. For FAA Use Only

4. Type		5. Unit Identification			
Repair	Alteration	Unit	Make	Model	Serial No.
<input type="checkbox"/>	<input type="checkbox"/>	AIRFRAME	_____	(As described in Item 1 above)	_____
<input checked="" type="checkbox"/>	<input type="checkbox"/>	POWERPLANT	Pratt & Whitney	JT8D-17	689892
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER			
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type		
			Manufacturer		

8. Conformity Statement

A. Agency's Name and Address		B. Kind of Agency		C. Certificate No.	
Name Turbine Engine Solutions, Inc		U. S. Certificated Mechanic		Manufacturer	
Address 13570 SW 129th Street		Foreign Certificated Mechanic			
City Miami State Florida		<input checked="" type="checkbox"/> Certificated Repair Station		Q6GR293Y	
Zip 33186 Country USA		Certificated Maintenance Organization		Limited Powerplant	

D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

Extended range fuel per 14 CFR Part 43 App. B <input type="checkbox"/>	Signature/Date of Authorized Individual Pablo Acosta 04/15/2008
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7. Approval for Return to Service

Pursuant to the authority given persons specified below, the unit identified in Item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is Approved Rejected

BY	FAA Fit. Standards Inspector		Manufacturer	Maintenance Organization	Persons Approved by Canadian Department of Transport
	FAA Designee	<input checked="" type="checkbox"/>	Repair Station	Inspection Authorization	Other (Specify)

Certificate or Designation No. Q6GR293Y	Signature/Date of Authorized Individual Pablo Acosta 04/15/2008
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NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

[Empty box for Nationality and Registration Mark]

04/15/2008

Nationality and Registration Mark

Date

Customer: Avlition Technologies

Engine Serial Number 688892 JT8D-17

Engine Total Time: 55,264 TSHS: 0

Engine Total Cycles: 28,303 CSHS: 0

Reason For Shop Visit: C/W ASB 6431, Hot Section & Repairs As Necessary.

01. Performed receiving inspection and video Borescope inspection.

02. N-1 Compressor Module:

C-1 Hub & Blade Assy. was removed cleaned, inspected, 7ea blades were replaced with an overhauled parts.

C-2 Disk and Short Stack rotor components were not removed.

C-1 Hub was static balanced and reinstalled.

03 Intermediate Case Module:

The module was not disassembled; inspected for continue time only.

1ea Compressor Bleed Valve was overhauled.

04. High Pressure Compressor Module:

The N-2 module was overhauled.

Disassembled, cleaned and inspected for corrosion.

Rotor components, reworked or replaced as necessary, static balance, reassembled and dynamic balanced.

Complied with AD 2003-12-07 IAW ASB 6431 R1.

Complied with AD 2006-17-07 IAW ASB 6430/6468.

05. Diffuser Case Module:

The Diffuser Case Assy. was inspected for continue time.

Fuel Nozzles & Supports assembly were bench checked.

2ea Compressor Bleed Valves were overhauled.

The remaining parts were inspected for continue time

06. Hot Section Module:

Hot section inspection accomplished.

Overhauled combustion chambers Cat 2A installed: P/N 780853 7 ea. & P/N 780854 2ea.

Complied with AD 86-09-02 R2 IAW ASB 5639 R10 & ASB 5639.

Nozzle Case Assy was inspected and repaired as necessary.

Inner & outer duct were installed in overhauled condition.

46ea 1st Stage NGV installed in Overhauled condition P/N 798771

Installed overhauled T-1 Outer Air Seal P/N M2355

07. High Pressure Turbine Module:

The module was fully disassembled.

All components were cleaned, inspected and repaired as necessary.

10ea T-1 blades were replaced with overhauled parts P/N: 794701

Complied with AD 2005-25-05 on T-1 Disk

The remainder of the components were inspected and repaired as necessary.

The module was reassembled and semi-dynamic balanced.

Additional Sheets Are Attached

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8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

	04/15/2008
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Nationality and Registration Mark Date

Engine Serial Number 689892 JT8D-17

08. Low Pressure Turbine Module:

The turbine was disassembled, cleaned and inspected.
T-3 and T-4 Disks were Overhauled P/N 774403
T-3 and T-4 Blades were Overhauled P/N 629704
The remaining components were inspected for continue time.
The turbine was static balance, reassembled and dynamic balanced.
Complied with AD 2005-25-05 on T-3 and T-4 Disks.

09. Exhaust Case Module:

The module assembly was inspected for continue time.

10. Main Accessory Gearbox Module:

The accessory gearbox assembly was removed, opened, inspected and reassembled.
Pressure check performed.

11. Accessories:

All accessories were inspected for continue time and functionally check at Test Cell.

12. The engine was reassembled and test cell run was performed IAW P&W JT8D E/M P/N 481672 rev.165 test #3, engine passed test within acceptable limit.

13. Performed post-test video Borescope inspections.

14. The engine fuel system was long term preserved for 90 days IAW P&W JT8D E/M 72-00-00 servicing -04.

Note: All work performed IAW JT8D E/M P/N 481672 REV. 165 4/15/08

Note: Original paperwork was sent to the customer, a copy is on file at Turbine Engine Solutions, Inc. under Work Order number 1176

..... NOTHING FOLLOWS

Additional Sheets Are Attached